

Parameter	Feature	Comment
<b>Spacecraft launch mass</b>	~310 kg	Payload inclusive
<b>Spacecraft size</b>	Ø 1,500 x 1,940 mm height	Hexagonal shape
<b>Attitude subsystem</b>	Pointing error: 0.15° (3σ) Attitude knowledge: 0.1° (3σ) Stability: 0.009°/s (3σ)	3-axis stabilization
<b>Agile platform</b>	±45° pointing in cross-track (max) ±30° pointing in cross-track (nominal)	High-performance AOCS for pointing accuracy & stability
<b>OBC (On-Board Computer)</b>	LEON3 FT, with VxWorks OS	Fully redundant, cross-strapped C&DH, 2 CAN busses (500 kbit/s each, redundant)
<b>Power generation</b>	450 W @ EOL Li-Ion battery, capacity of > 30Ah @ EOL	4 GaAs solar panels
<b>GSD (Ground Sample Distance)</b>	1 m (PAN), ~ 4 m (MS)	@ 600 km altitude, nadir view
<b>Swath width</b>	> 12 km	
<b>Spectral bands</b>	PAN + 4 MS	PAN + RGB, NIR
<b>TT&amp;C</b>	S-band (32 kbit/s in up- and downlink)	Telemetry, telecommand & tracking, CCSDS protocol, PTD chip, authenticated commands and encrypted TLM
<b>Image transmission</b>	X-band with 160 Mbit/s downlink rate	QPSK modulation
<b>On-board recording capability</b>	256 Gbit	Equivalent to 1400 km observation strip
<b>Propulsion subsystem</b>	> 10 mN of thrust, > 1000 s of I <sub>sp</sub> , 3 kg of Xenon	Orbit control and maintenance
<b>TCS (Thermal Control Subsystem)</b>	Passive and and thermal control	5 operational heaters+2 survival heaters; dedicated temperature sensors, uniform distribution